

Europäisches Patentamt  
European Patent Office  
Office européen des brevets



(11) EP 1 231 046 A2

(12)

## EUROPEAN PATENT APPLICATION

(43) Date of publication:  
14.08.2002 Bulletin 2002/33

(51) Int Cl.<sup>7</sup>: B29C 70/34, B29C 70/44

(21) Application number: 02380034.5

(22) Date of filing: 13.02.2002

(84) Designated Contracting States:  
AT BE CH CY DE DK ES FI FR GB GR IE IT LI LU  
MC NL PT SE TR  
Designated Extension States:  
AL LT LV MK RO SI

(30) Priority: 13.02.2001 ES 200100317

(71) Applicant: Eads Construcciones Aeronauticas  
S.A.  
28022 Madrid (ES)

(72) Inventors:

- Nogueroles Vines, Pedro  
28022 Madrid (ES)
- Garcia Garcia, Aquilino  
28022 Madrid (ES)
- Cerezo Pancorbo, Carlos  
28022 Madrid (ES)

(74) Representative: Elzaburu Marquez, Alberto et al  
Elzaburu S.A.,  
Miguel Angel, 21  
28010 Madrid (ES)

(54) Method for manufacturing elements of composite materials by the cobonding technique

(57) Method for manufacturing elements made from composite material using the co-bonding technique, in which uncured elements (preformed beams (2)) are bonded onto another cured element (the skin (3)) with multiple thickness changes, using a rigid tooling made of invar used for support and positioning during the curing process. Each element is made with preimpregnated material using automated tape laying. The preform of the beams (J-section) is obtained by hot forming of flat

laminates. The final curing and bonding to the precured skin (co-bonding) is performed using a direct vacuum bag in an autoclave. A flat development of the vacuum bag is performed, it is traced with a numerical control machine and it is made prior to being placed on the tool. For large surfaces with difficult access the final adjustment is performed with the tool and the part in a vertical position, due to the ergonomic difficulties involved in working on certain areas of same. The invention is applicable to the field of aeronautics.

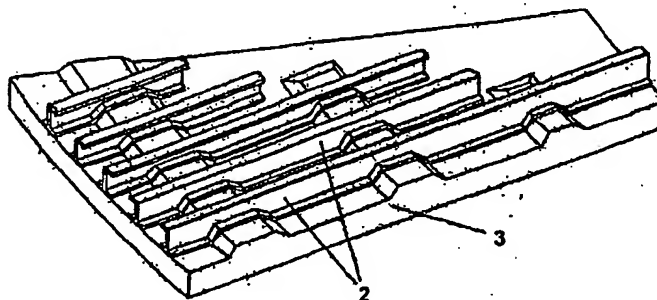


Fig. 1

EP 1 231 046 A2

**Description****FIELD OF THE INVENTION**

[0001] The present invention relates in general to methods for manufacturing a composite material structure, in which on a precured element (skin) are glued other uncured elements (beams) by placing an adhesive layer between them (between the precured element and each uncured element), thereby obtaining a structural union. The adhesive is cured at the same time as the latter elements.

[0002] More specifically, the object of the invention is to develop the required theoretical concepts and the corresponding manufacturing methods for providing a union system by co-bonding of one or several elements (beams) made from composite materials and uncured, and a base (skin) also made of composite material but which is precured, with multiple changes in thickness. A precise adjustment must be obtained of the uncured elements, both with the adhesive surface (skin) and with the other, upper surface.

[0003] For this purpose, the tooling used is the most relevant factor, which is in this case a rigid invar rod (described in detail below) with a direct bag that allows to obtain a high dimensional precision at the same time as a tight positioning tolerance. To clarify the term "direct bag", it should be pointed out that the direct vacuum bag concept relates to the fact that the elements comprising the vacuum bag (FEP or fluoro-ethylene-propylene, AIRWEAVE type aerator and bag plastic) are directly on the part to be cured without any interposed tooling. This ensures a uniform consolidating pressure.

[0004] The union is achieved by curing the adhesive layer under strict pressure conditions and at its polymerization temperature, which must match that of the resin of the uncured elements as both chemical processes take place simultaneously in the same autoclave cycle.

[0005] Likewise, the union effected is designed to withstand shear loads applied to the skin by the beams, due to deflections of the structure, and detachment forces applied on the beams by the skins, as well as various types of internal pressures such as those of a fluid when the torsion box is the fuel tank.

**BACKGROUND OF THE INVENTION**

[0006] The most remarkable characteristic of the present invention is the use of a rigid tooling (a system of rigid tools and rakes) for the bonded union, combined with applying autoclave pressure using a scheme with a vacuum bag in direct contact with the elements to be bonded and cured.

[0007] In order to bond the uncured elements to a precured skin which must match another complex surface at the unbonded end, a manufacturing system was initially developed with a flexible tooling using the "inflatable tool" technique. These tools were made from an

elastomer material stiffened as required with carbon fiber.

[0008] The high cost and low reliability of this tooling spurred the development of a rigid tooling system to solve these problems; this is the co-bonding system with rigid tools.

During the development stage of the rigid tooling trials were performed with tools of various configurations:

■ Several configurations were tested with steel material, which were discarded because of the thermal gradients generated which resulted in deformations of the part to the point of not obtaining the required quality.

■ Two constructive solutions have been tested using invar:

- Rigid tools made from welded sheets which are later machined. This solution is the lightest but its construction is extremely complex and involves several deformation and straightening operations during fabrication.

It is also possible to leave a small wall thickness after machining, with the resulting risk of collapse of the tool in the autoclave. The resulting weight does not allow manual handling.

- Rigid tools made from a sheet with a sufficient thickness and enlightened by machining, and later covered by a welded plate.

[0009] The enlightened material weighs  $\approx 25$  kg as compared to a weight of the solid tool of around 150 kg. This enlightening is not justified due to handling issues as it greatly increases the tool fabrication cycle and its handling still requires additional means.

[0010] As well as the use of different materials and configurations of the rigid tools, another basic aspect in the use of this type of blade-shaped rigid tooling is the distance between the edge of the rigid tool and the radius of the beam foot. The following configurations were tested:

- The rigid tool extending 2 mm into the radius.

- The rigid tool remaining 2 mm above the radius.

- The rigid tool extending as far as the middle of the radius.

[0011] It was concluded that the rigid tool should end above the radius of the foot, as this configuration provides the best dimensional and quality results, as well as facilitates demolding.

[0012] Later studies led to an optimization of the distance between the rigid tool and the beam foot radius,

arriving at the conclusion that the ideal distance was 3 mm from the edge of the rigid tool to the start of the beam foot radius.

[0013] The results obtained indicate that rigid tools should be made of solid invar, as this simplifies their construction and improves dimensional tolerance. Additionally, they are handled in all cases with auxiliary means and not manually, regardless of their configuration.

[0014] As regards the bonded unions, using a different type of tooling, the prior art closest to the application are those relating to:

1. Joining beam stiffeners of the torsion box for the A330-340 airplane horizontal stabilizer (currently in the production stage).
2. Joining the longitudinal stiffeners for the skin of the torsion box of the CASA 3000 airplane wing (in prototype stage).
3. Joining auxiliary longitudinal beams to the skin of the torsion box of test FB.5-1 of the technological development program for large airfoils (GSS) to be applied to the horizontal stabilizer of the A3XX.

[0015] From the results of the above experiences and from other relevant manufacturing studies and tests it was concluded that the application of the method of the present invention is feasible and reliable for its use in parts of highly demanded withstanding structures and with high quality requirements, with complex shapes and strict dimensional tolerances.

#### **FIELD OF APPLICATION OF THE INVENTION**

[0016] This invention is applicable to the manufacture of structures made of composite materials in which participate a precured element (skin) and other uncured elements (beams) that are cured simultaneously to their union to the precured element.

[0017] The structures for which this technique would be applicable are such as:

- Airplane structures and controls, such as airfoils, moving airfoil surfaces, fuselages.
- Space ships
- Marine and land vehicles
- Industrial machinery and equipment.

The various manufacturing stages which comprise the full process are:

#### **Fabrication of the skin**

[0018]

- Tape laying on a curved tool.
- Placing the vacuum bag on a laminate.
- Curing in an autoclave.

- There is no demolding operation nor a non-destruction inspection.

#### **Fabrication of the J-beams**

[0019]

- Flat tape laying.
- 2D cutting in fresh state on patterns.
- Mounting patterns until final configuration of the beam cloths.
- A first hot forming cycle to obtain two L-shaped beam halves.
- Placing one half on the other.
- A second hot-forming cycle, to provide the final J-shaped beam.
- 3D cutting of the uncured beam rises as well as other cutting to obtain the final size of the beam after the curing cycle.

#### **Fabrication of the vacuum bag**

[0020]

- Approximate flat layout of the final bag configuration.
- Tracing the bag in a flat machine with numerical control or manually with jigs or Mylar. The position of the beams and fasteners on the radii is traced.
- Formation and manual attachment of the fasteners.

#### **Fabrication of the final structure: co-bonding**

[0021]

- Assembling the beams on rigid invar tools on auxiliary preassembly benches. Each bench has two rigid tools to allow ergonomic working conditions.
- Placing all possible elements of the final vacuum bag on the beams in the preassembly benches. Additionally, a consolidation is carried out to ensure adjustment on the skin. For this, the preassembly benches are provided with a surface which perfectly resembles the surface of the skin.
- Transfer of rigid tools + rakes + beams to their final position on the skin.
- Placing the remaining elements of the vacuum bag.
- Assembling the prefabricated and checked vacuum bag.
- Final adjustment of the vacuum bag with the assembly in a vertical position for large surfaces with difficult access to certain areas.
- Autoclave curing cycle.
- Demolding.
- Non-destructive inspection of the skin.
- Re-edging (only for the skin as the system of rigid tools allows to obtain beams with their final geometry).

- Non-destructive inspection of beams.
- Priming and painting.

#### **Materials**

[0022] The materials to be used will be composite materials, in which the fibers and resin can be:

#### **FIBERS**

[0023]

- Carbon fiber.
- Glass fiber.
- Ceramic fiber.
- Aramid fiber.
- Boron fiber.

#### **RESINS**

[0024]

- Epoxy resin.
- Thermoplastic resin.
- Other thermosetting resins.

#### **SUMMARY OF THE INVENTION**

[0025] The object of the invention is a method for manufacturing composite material structures in which several uncured elements (beams) are joined to a precured element (skin) so that the union has structural requirements.

[0026] The bonding and curing of the beams is achieved by a prior forming and a final curing in an autoclave with a direct vacuum bag.

[0027] The uncured elements have a J-shaped cross section.

[0028] The basis of the manufacturing method is the optimized design of forming tools (made of aluminum and improved wood with an integrated vacuum system for overturning) and particularly curing in an autoclave, rigid tools made of invar (to avoid deformations due to thermal expansion) and the automation of all processes.

[0029] The method is applicable to any base structure which must be stiffened by elements with a precise geometry.

[0030] The tape laying technique can be either manual or automated, although the automated tape laying system optimizes the process considerably.

[0031] In a specific embodiment the invention discloses a method for manufacturing precured parts of composite material by using uncured J-beams, in which are structurally joined at least two parts made of composite materials, of which a first part known as the base part or skin is in a cured state and a second part or parts, known as beams, are uncured, and in which the two parts are joined by a layer of structural adhesive so that

the second part is compacted against the first, with a suitable cross linking of the resin of the composite material, and so strongly bonded to the skin of the first part that the required strength of the adhesive layer is ensured. This method is characterized by the following stages: laminating superposed layers of preimpregnated composite material so that the fiber orientation is adapted to the structural requirements of the part to be obtained, obtaining from the resulting laminates on one hand the base part and on another a set of basic stacks used to form the second part; curing the base part in an autoclave; cutting the flat laminate with the areas of different thickness from which the second parts are obtained; assembling packages from the patterns obtained in the previous cutting; hot forming in two cycles, by applying heat and vacuum, of the previously obtained flat configurations to obtain a preform with a J-shaped cross-section; mounting the preforms on the curing tools on auxiliary preassembly benches which simplify this task; precise positioning of all tooling (rigid tools + rakes) and J-shaped parts on the precured base; mounting a previously made and checked vacuum bag; overturning the part and the tool to a vertical position when the parts have a large area and are difficult to access, in which position the fine adjustment of the vacuum bag is performed; and performing the autoclave curing cycle.

[0032] In accordance with the invention a base part and one or more second parts are joined to obtain a finished precured part. The uncured elements to be bonded are obtained from flat laminates of varying thicknesses in some areas, which are later cut and stacked in packages until the final configuration of the part, with packages of at least two cloths being stacked and in no case with two cloths touching each other.

[0033] Likewise, the uncured elements to be bonded are hot formed to obtain preforms with the final geometry, so that they can be easily mounted on the curing tools (rigid tools). The hot forming tools are made of aluminum with improved wood on their top part, which is in contact with the fiber, in order to prevent heat transfer losses as well as in the integrated vacuum system for overturning said tools.

[0034] In addition, the curing tools generally have a rectangular trapezoid cross-section so that the geometrical quality of the part is ensured, allowing to adjust the beams on their top surface with another part of the type of the base part. These curing tools are made of invar to prevent deformations due to thermal expansion during the autoclave cycle.

[0035] Furthermore, between the edge of the rigid tool and the foot radius of the beam there is a 3mm separation which ensures the geometrical quality of the part as well as facilitates the demolding; the autoclave curing process is performed at a pressure between 585 kPa and 896 kPa, and at a temperature of up to 190°C depending on the composite material used, with a heating gradient of 0.5 to 2°C/min.

[0036] With the method of the invention parts are ob-

tained that can be applied in structures and controls of aerospace, marine and land vehicles, as well as in industrial machinery and equipment. Specifically, the base part (skin) comprises the skin of an airplane wing, a stabilizer or any other element which must be stiffened to fulfill its structural functions.

[0037] In accordance with the invention, the uncured parts have a J-shaped cross section and thickness between 1 and 6 mm, while the base part has a length of up to 7 m and is shaped as a delta.

[0038] The vacuum bag used in the method of the invention is quite large, so that it is traced with a numerical control machine and made before it is placed.

[0039] The composite material used in the method of the invention consists of fibers and resins selected among glass fiber, carbon fiber, aramid fiber, boron fiber, epoxy resin, thermoplastic resin and other thermosetting resins.

#### **BRIEF DESCRIPTION OF THE DRAWINGS**

[0040] The above and further characteristics and advantages of the invention will become apparent by means of the following description made with reference to the accompanying drawings, where:

Figure 1 is a perspective view of a structure made of composite material obtained by the method of the invention.

Figure 2 is a perspective view of a rigid invar tool used in the method of the invention.

Figure 3 is a typical cross section of a rigid invar tool such as that of figure 2.

Figure 4 shows the rigid tool of figure 2 and a J-beam attached to it enclosed in a direct vacuum bag.

Figure 5 shows a J-beam attached to a rigid invar tool such as that of figure 2.

Figure 6 shows the successive steps of the manufacture of a preform for a J-beam in accordance with the method of the invention.

Figure 7a-c shows enlarged sectional views of the two hot forming tools used in the manufacturing method illustrated in figure 6.

Figure 8a-b shows two methods for attaching a rigid invar member to a support rake.

Figure 9 shows finally a schematic representation of the relative positioning of the various elements required to carry out the method of the invention.

#### **DETAILED DESCRIPTION OF THE INVENTION**

[0041] The manufacturing method of the invention applicable to composite materials is associated to a set of various prior processes which allow an optimized co-bonding.

[0042] The basis of the success of the method is the tooling used. The method and system of rigid tooling, rigid tools (1), has been specifically conceived for peculiar geometries in which the J-section beams (2) must be bonded to a skin (3) with multiple thickness variations, and where the beams must match well another surface on the upper side thereof.

[0043] As the beams (2) must maintain a given height with a tight tolerance the rigid tools (1) must be completely still during the curing cycle, without exerting any compacting pressure against the beam foot, which makes the process even more complicated.

#### **20 FABRICATION OF THE SKIN**

[0044] The base skin is made by automatic tape laying (ATL), although the tape can also be placed manually, in both cases using unidirectional tape with a 150 mm band width. A relatively low width is used to allow laminating considerably curved surfaces with large local differences in thickness, due to the local application of structure loads in its service lifetime.

[0045] To prevent the problems with the different thermal expansion between the tool (4) and the material used (carbon fiber tape) an invar tool is used, as this material has a thermal stability which is quite similar to the carbon fiber laminates.

[0046] The curing cycle is performed at a high pressure (896 kPa) to ensure that no pores appear inside the element. Without demolding the skin (3) from the tool (4), the co-bonding operation of the beams (2) is performed after their preform is obtained.

#### **40 FABRICATION OF THE J-BEAMS**

[0047] The J-beams (2) are made using the package technique in order to maximize the automation of their manufacture.

[0048] In order to avoid having to place the cloths one by one, a flat laminate is automatically tape-laid with the correct orientations and different thickness that vary from 0.5 mm (2 layers of carbon fiber tape) to 1.75 mm (7 layers of carbon fiber tape). Flat tape laying is with unidirectional tape with a 300 mm width to shorten machine use times (or manual tape laying times).

[0049] This laminate is taken to the panex 2D cutting machine (for uncured cutting), from which the patterns are obtained that will be part of each beam (2) (the term "panex" refers to an uncured stack of carbon fiber tape and/or fabric). Before hot forming the packages are mounted until the required cloth sequence is obtained. Some of the mounted patterns are full for the entire part

and others are local reinforcements for certain areas.

[0050] The preform of the J-beam (2) is obtained by two hot forming cycles shown in figures 6 and 7.

[0051] In the first cycle the foot (5, 5') of each half of the beam (2) is bent to obtain two L's. The tool (6) is laid on the tool (6'), with the laminate held onto the tool by the integrated tool vacuum system. An end strip (7) of carbon fiber (2 layers) and a layer of adhesive (8) are placed on the foot of the beam, without removing the support paper to prevent the beam from adhering on the preassembly benches as explained further below.

[0052] The central gap left in the foot between the two halves of the beam is filled with a part (9) of carbon fiber (rowing) with a triangular section and orientations from (+/-) cut by the numerical control machine.

[0053] In figures 6 and 7 can also be seen a holder (10), a pin (27) for attaching the former, an FEP sheet (11), two adhesive tapes (12) and two adhesive angles (13, 13').

[0054] In the second cycle the final J is obtained by folding the end (the head) of the beam (2) distant from the foot (5, 5').

[0055] Stages a) and b) shown in the figure 7 correspond to the first cycle described and stage c) of figure 7 corresponds to the second cycle described above.

[0056] The tools (6, 6') are made of aluminum and improved wood in contact with the laminates in order to minimize the heat transfer from the laminate to the tool. During forming the vacuum must be created slowly (10mmHg/min) to allow the cloths to slide on each other preventing the appearance of wrinkles at the turning radii. The vacuum is applied when the laminate is at the right temperature (between 45 and 60°C).

[0057] In order to avoid the re-edging operation for beams (2) after they are joined to the skin (3), which is complex due to their final geometry, the longitudinal rises and the head rises are cut when uncured, as well as any orifices or holes in the beam core using a 5-axis 3D numerical control machine. After this operation the beam has its net size.

#### VACUUM BAG

[0058] For large and complex shapes of the vacuum bag (14) (figure 4) a bag prefabrication method has been developed which comprises the following stages:

- Approximate flat development of the final bag configuration.
- Tracing the bag in a 2D numerical control machine or with jigs or Mylar. The position of the beams and fasteners on radii are traced.
- Manual creation of fasteners.
- Checking the vacuum bag and closing it on itself.
- Storage, suitably protecting the plastic until it is used.

[0059] For large surfaces an overturning system is

used to mount the vacuum bag, with tool (4) taken to a vertical position which allows access to its inside. During the bonding process it is necessary to access the entire surface in an ergonomic manner in order to carry out a thorough adjustment and visual inspection of the complex vacuum bag used, before the autoclave cycle. The adjustment of the bag and tool or bag and part determines the reliability of the bag and the correct shape of the part. For this purpose the entire bag-tool unit is raised to a position which will later allow to rotate it to a vertical position.

[0060] The structure of bag (14) consists of a two-sided tape (15), a separator (11), not shown in figure 4 and which corresponds to the aforementioned FEP sheet (11), an AIRWEAVE type aerator (16) and an external plastic film (17), such as of nylon.

[0061] The abbreviation TIP of figure 4 indicates "typical", as applied to a measurement.

#### CO-BONDING PROCESS

[0062] The following basic tooling is used for the co-bonding process:

- Rigid tools
- Rakes
- Preassembly benches

#### RIGID TOOLS

[0063] These are the main part of the tooling and their adjustment and positioning determines the dimensional quality of the co-bonded part.

[0064] They are the support and positioning elements for the beams (2) during the curing cycle, and at the same time serve to precisely position each beam. A typical section of a rigid tool (1) is schematically represented in figure 5.

[0065] The step (18) on the upper part is designed as a stop for the head (19) of the beam (2), although a 1 mm clearance is provided to allow the resin to overflow during the curing process.

[0066] In its lower part the profile of the rigid tool (1) ends as a sharp edge (20), for several reasons:

- to allow a perfect coupling of the vacuum bag (14) without any bridging of it, which would cause the bag to break or the beam (2) to be deformed in that area;
- to prevent resin from overflowing in that area, which would greatly difficult demolding as well as result in deformations of the radius of foot (5, 5') of the beam (2).

[0067] The edge (20) ends 3 mm above the start of the radius of the foot (5, 5') in order to allow the demolding operation.

[0068] The edge (20) of the rigid tool (1) will accurately

follow the shape of the skin (3) with raised and lowered areas due to thickness changes.

[0069] With this design for the tool the foot (5, 5') of the beam (2) (the area to be bonded) is subjected to the direct pressure of the autoclave through the vacuum bag (14).

[0070] Those areas of the rigid tool (1) where there is no beam (2) are covered by metal elements (28) (fillers) which simulate the thickness and the foot (5, 5') of the beam (2), so that the bag (14) does not slip into the bottom slit which would be left with the resulting risk of breaking same.

[0071] As mentioned above, the rigid tools (1) are made of invar in order to prevent the deformation of the beam (2) during the curing cycle as a result of the different thermal expansion of the beam and the rigid tool.

[0072] At the longitudinal ends of the rigid tools (1), just at the end of the beam (2), are made grooves (21) for inserting retainers (29) of airpad or Teflon, which can be easily disassembled during demolding and which allow to retain the excess resin from beams (2) during the autoclave cycle.

## RAKES

[0073] These are the supporting elements for the rigid tools (1). Some are placed on the tool (4) of the skin (3) and others on the cured skin (3), simply resting on it. Their purpose is to ensure the accuracy in height and verticality of the rigid tools, while also ensuring its longitudinal alignment. A typical scheme for a rake (22) is shown in figures 8a-b, where figure 8a shows the attachment of rigid tools to rakes in the section of the engagement of the wing to the fuselage, while figure 8b shows the attachment of rigid tools to intermediate and wing tip rakes.

[0074] The gaps between the rigid tool (1), rake (22) and skin (3) must be avoided in order to minimize the risk of bag (14) breaking. This is extremely important in the design of the final co-bonding tools.

[0075] The vertical support surfaces of the rigid tools (1) are provided with a small cemented plate 2 mm thick to prevent their wear. On these are placed shoes (23) which serve two functions:

- to give the rigid tool (1) the correct height
- to allow the demolding operation, for which with the rigid tool (1) secured in place the shoes (23) are removed and the rigid tool is moved 2mm towards the skin (3), a sideways motion is effected to release the head (19) of the J part of the beam (2) and finally it is raised to carry it to the corresponding storage cart.

[0076] The rigid tools (1) are attached to the rakes (22) by screws (24) and clamps (25) as well as with straps (26).

[0077] Figure 2 shows the cooperation of the rigid tool

(1) and the rakes (22) for supporting the beam (2). In this figure the reference (30) represents the area of assembly of the rake on the wing tip, with (31) representing the area of assembly of the fillers (28), (32) representing the area of assembly of the rake in the engagement of the wing to the fuselage and (33) being the area where the beam is housed.

[0078] Figure 9 shows an enlarged view of the relative position of the rigid tool (1), the rake (22), the filler (28), the shoe (23), the retainer (29) and the beam (2), with the shaded area representing a typical cross section of the rigid tool (1).

## PREASSEMBLY BENCHES

[0079] The assembly of the beams (2) on the skin (3) is performed on auxiliary preassembly benches whose surface reproduces the theoretical surface of the skin. Each of these benches holds two beams with the corresponding rigid tools (1) and rakes (22). This allows an ergonomic working position.

[0080] The rakes (22) are mounted on the preassembly bench, the rigid tools (1) are placed on the rakes and then the beam is placed on the rigid tool, where it is consolidated. The foot (5, 5') of the beam includes the adhesive strip with the support paper attached to prevent it from adhering to the bench. Finally, the assembly is transported to the skin (3) where it is precisely positioned.

[0081] After the entire tooling and the beams (2) are positioned on the skin (3), the scheme of the vacuum bag (14) is completed by a visual inspection of the full tightness.

[0082] For large surfaces, the final adjustment of the bag (14) will be made using the tool and the part in a vertical position.

[0083] After the vacuum bag is formed it is inserted in the autoclave for curing and bonding of the beams (2) to the skin (3).

[0084] The following operations are summarized below:

- Demolding: as explained above.
- Automated ultrasound inspection of the skin (3).
- Re-edging in a 5-axis numerical control machine.
- Automated and manual ultrasound inspection of the beams (2) and the re-edged areas.
- Priming and painting.

[0085] In the above the essential characteristics of the invention have been described, although it is understood that it would be possible to change certain details of the fabrication method and tooling as developed by the applicant. For this reason it is intended that the scope of the invention be limited only by the contents of the appended claims.

# Claims

1. Method for manufacturing precured parts from a composite material with J-section beams applied in an uncured condition, in which are structurally joined at least two parts made from composite materials, of which a first part known as a base or skin (3) is cured and a second part or parts known as beams (2) are uncured, and in which the two parts are joined by a layer of structural adhesive so that the second part is compacted against the first with a suitable cross linking of the resin of the composite material, and so strongly bonded to the skin of the first part that the required strength of the adhesive layer is ensured, **characterized by** the following stages:
  - laminating superposed layers of preimpregnated composite material so that the fiber orientation is adapted to the structural requirements of the part to be obtained, producing from the resulting laminates on one hand the base part and on another a set of basic stacks used to form the second part;
  - curing the base part in an autoclave;
  - cutting the flat laminate with areas of different thickness from which the second parts are obtained;
  - assembling packages from the patterns obtained in the previous cutting;
  - hot forming in two cycles, applying heat and vacuum, of the previously obtained flat configurations to obtain a preform with a J-shaped cross-section;
  - mounting the preforms on the curing tools on auxiliary preassembly benches which simplify this task;
  - precise positioning of all tooling (rigid tools (1) + rakes (22)) and J-shaped parts on the precured base;
  - mounting a previously made and checked vacuum bag (14);
  - overturning the part and the tool to a vertical position when the parts have a large area and are difficult to access, performing the fine adjustment of the vacuum bag in this position; and performing the autoclave curing cycle.
2. Method as claimed in claim 1, **characterized in that** the base part and one or more second parts are joined to obtain a precured finished part.
3. Method as claimed in any of the previous claims, **characterized in that** the elements to be bonded in an uncured state are obtained from flat laminates with a varying thickness in different areas, which are later cut and stacked in packages until the final configuration of the part, stacking packages of at least two cloths and in no case placing one cloth against another.
4. Method as claimed in any of the previous claims, **characterized in that** the elements to be bonded in an uncured state are hot formed to obtain preforms with the final geometry, so that they can be easily mounted on the curing tools (rigid tools (1)).
5. Method as claimed in any of the previous claims, **characterized in that** the hot forming tools are made of aluminum with improved wood on their upper part which is in contact with the fiber, in order to prevent heat transfer losses, as well as losses of the integrated vacuum system used in the overturning operation of said tools.
6. Method as claimed in any of the previous claims, **characterized in that** the curing tools (1) have a section with a rectangular trapezoid shape so that the geometrical quality of the part is ensured, allowing to adjust the upper side of the beams (2) to another part of the base part type.
7. Method as claimed in any of the previous claims, **characterized in that** the curing tools (1) are made of invar to prevent deformations due to thermal expansion during the autoclave cycle.
8. Method as claimed in any of the previous claims, **characterized in that** between the edge (20) of the rigid tool (1) and the radius of the foot of the beam (2) there is a 3 mm separation that ensures the geometrical quality of the part, as well as demolding of same.
9. Method as claimed in any of the previous claims, **characterized in that** the autoclave curing takes place at a pressure between 586 kPa and 896 kPa and at a temperature of up to 190 °C, depending on the composite material used, with a heating gradient of 0.5 to 2°C/min.
10. Method as claimed in any of the previous claims, **characterized in that** parts are obtained that can be applied to structures and controls of aerospace, marine and land vehicles, as well as to industrial machinery and equipment.
11. Method as claimed in any of the previous claims, **characterized in that** the base part (skin (3)) consists of the skin of an airplane wing, a stabilizer or any other element that requires stiffening to fulfill its structural function.
12. Method as claimed in any of the previous claims, **characterized in that** the uncured parts have a J-shaped cross section.



13. Method as claimed in any of the previous claims,  
**characterized in that** the uncured parts have a  
thickness between 1 mm and 6 mm, and **in that** the  
base part has a length of up to 7 m, with a delta  
shape. 5
14. Method as claimed in any of the previous claims,  
**characterized in that** the vacuum bag (14) has  
large dimensions and for this reason it is traced in  
a numerical control machine and made prior to be- 10  
ing placed.
15. Method as claimed in any of the previous claims,  
**characterized in that** the composite material con- 15  
sists of fibers and resins chosen from among glass  
fiber, carbon fiber, aramid fiber, boron fiber, epoxy  
resin, thermoplastic resin and other thermosetting  
resins.

20

25

30

35

40

45

50

55

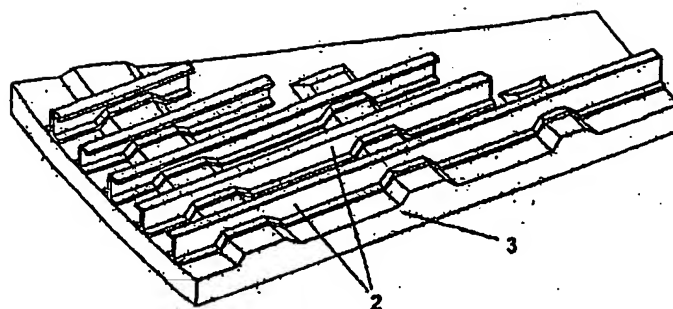


Fig. 1

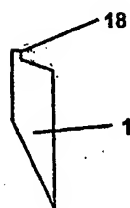


Fig. 3

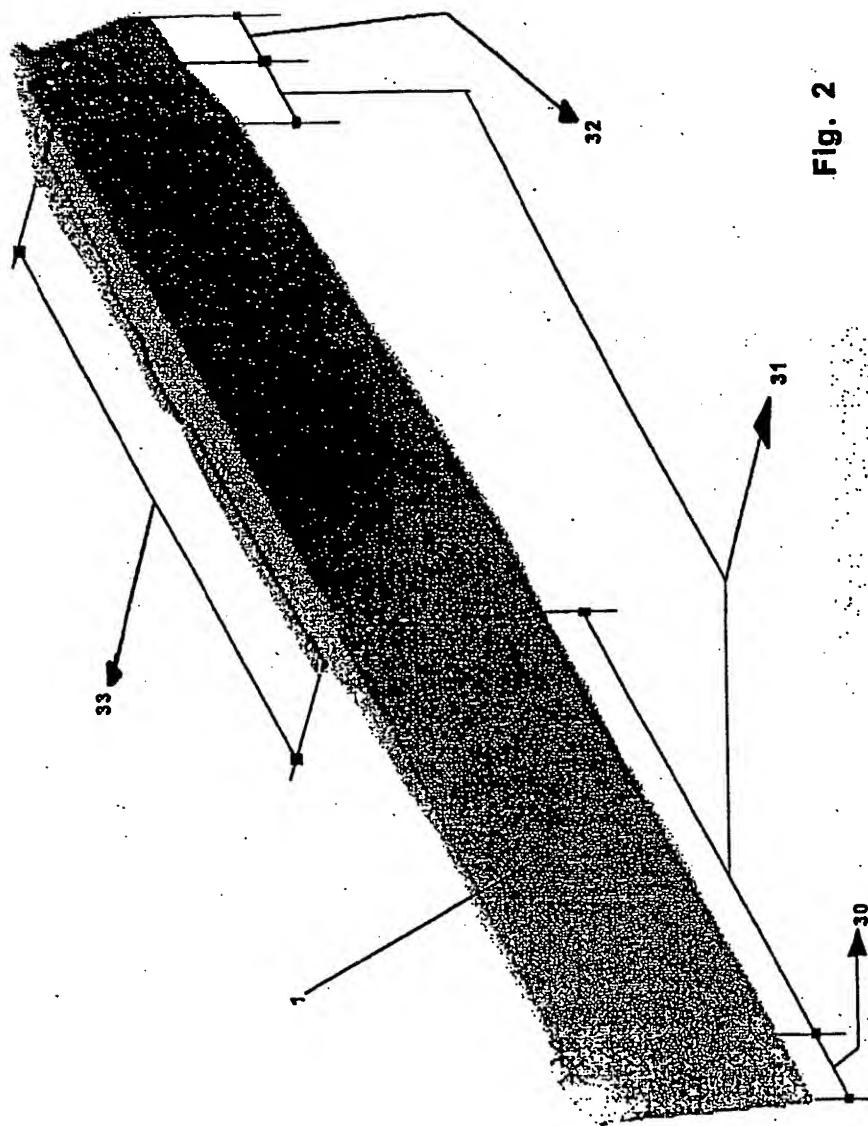
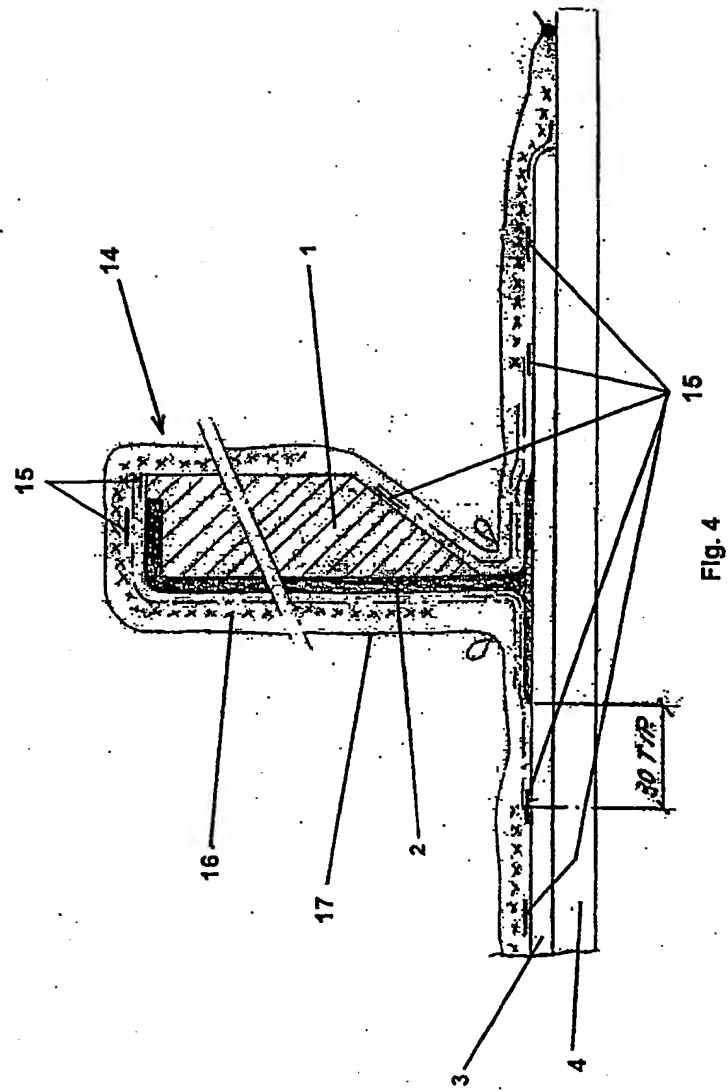


Fig. 2



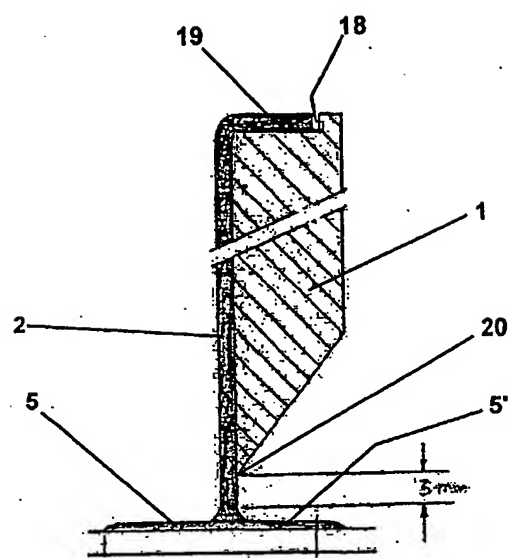


Fig. 5

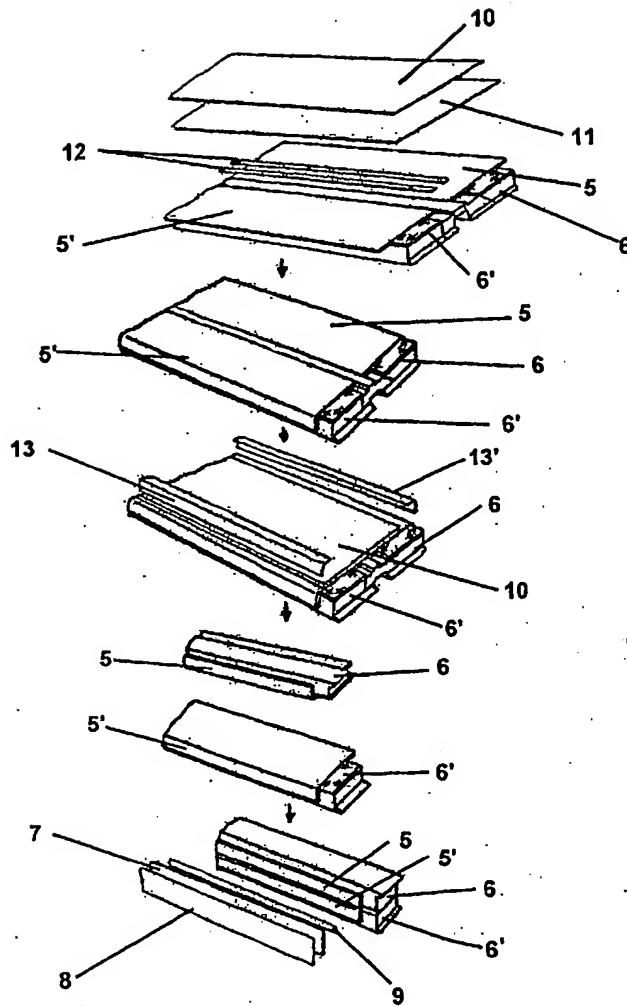


Fig. 6

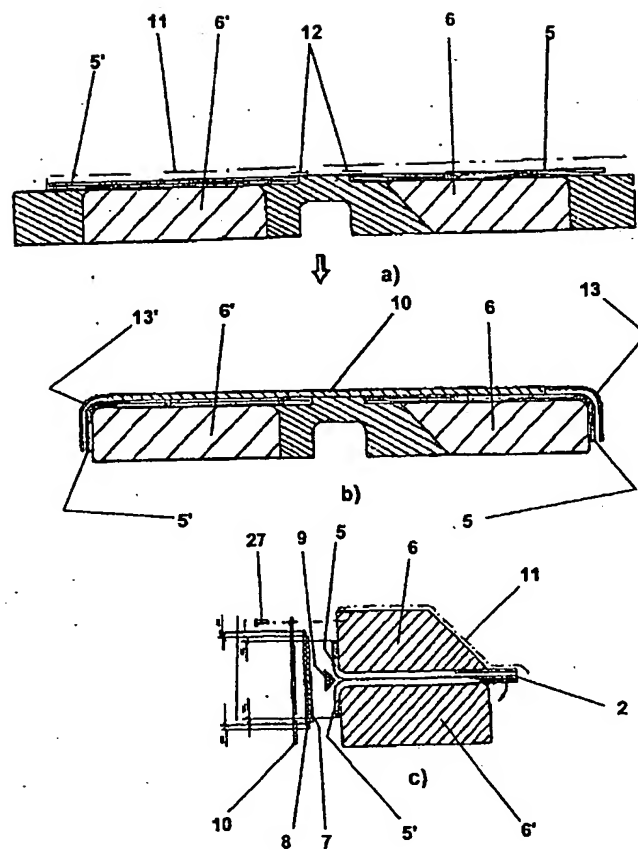


Fig. 7

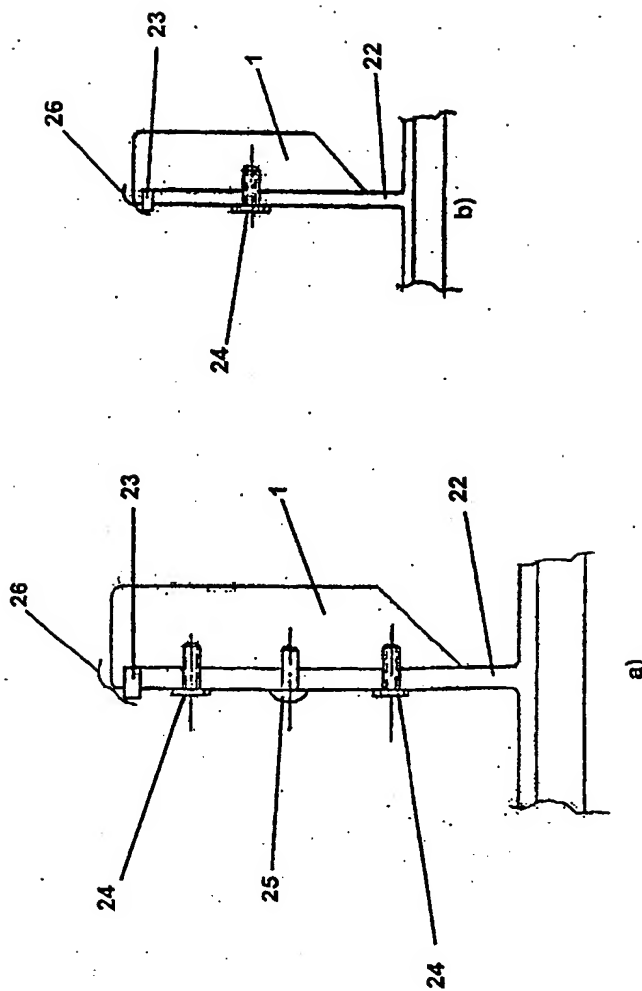


Fig. 8



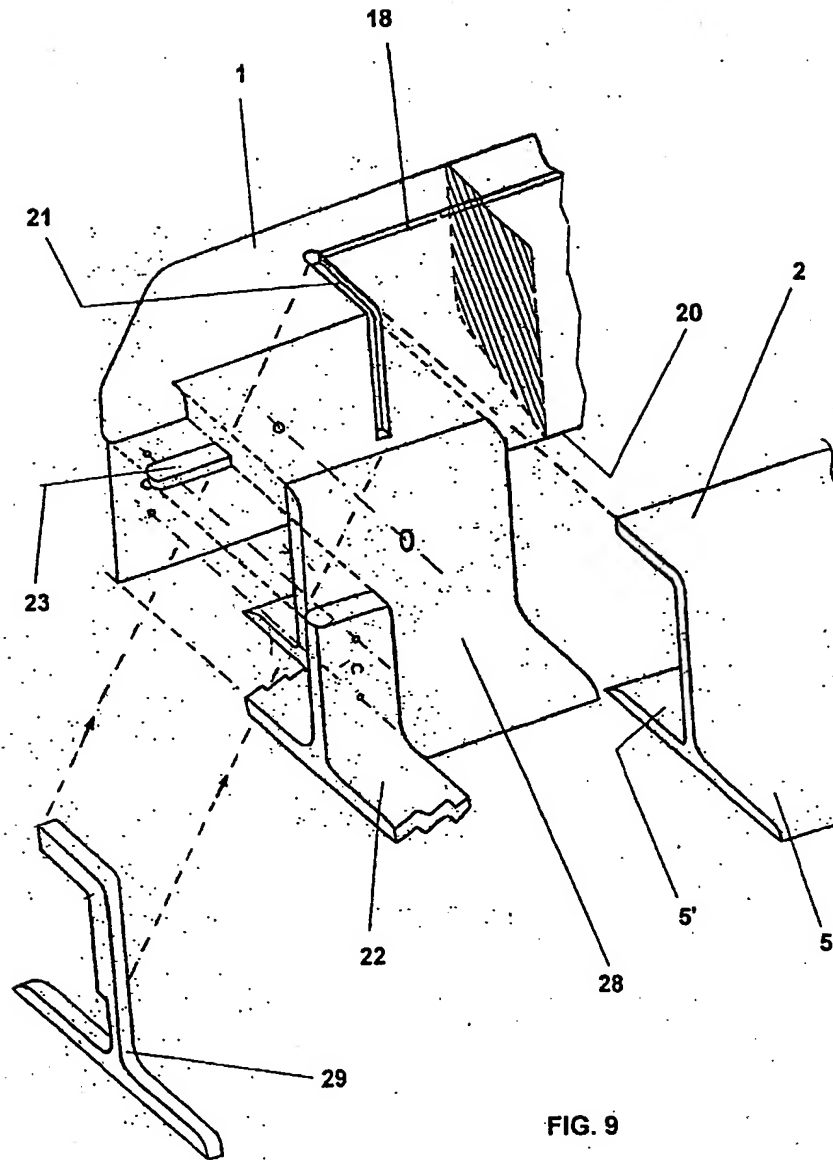
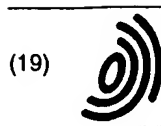


FIG. 9



Europäisches Patentamt  
European Patent Office  
Office européen des brevets



(11) **EP 1 231 046 A3**

(12)

## EUROPEAN PATENT APPLICATION

(88) Date of publication A3:  
30.10.2002 Bulletin 2002/44

(51) Int Cl.7: **B29C 70/34, B29C 70/44,  
B64C 3/18, B64C 3/24**

(43) Date of publication A2:  
14.08.2002 Bulletin 2002/33

(21) Application number: **02380034.5**

(22) Date of filing: **13.02.2002**

(84) Designated Contracting States:  
**AT BE CH CY DE DK ES FI FR GB GR IE IT LI LU  
MC NL PT SE TR**  
Designated Extension States:  
**AL LT LV MK RO SI**

(72) Inventors:

- **Nogueroles Vines, Pedro**  
28022 Madrid (ES)
- **Garcia Garcia, Aquilino**  
28022 Madrid (ES)
- **Cerezo Pancorbo, Carlos**  
28022 Madrid (ES)

(30) Priority: **13.02.2001 ES 200100317**

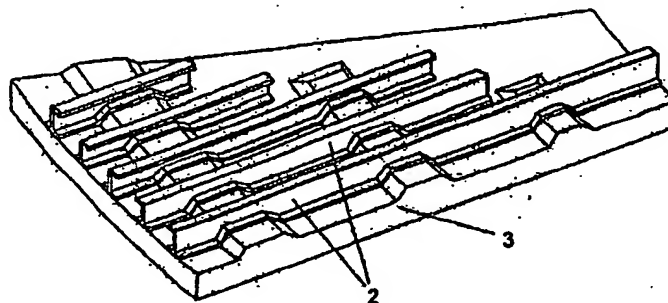
(71) Applicant: **Eads Construcciones Aeronauticas  
S.A.**  
28022 Madrid (ES)

(74) Representative: **Elzaburu Marquez, Alberto et al**  
**Elzaburu S.A.,**  
**Miguel Angel, 21**  
**28010 Madrid (ES)**

(54) **Method for manufacturing elements of composite materials by the cobonding technique**

(57) Method for manufacturing elements made from composite material using the co-bonding technique, in which uncured elements (preformed beams (2)) are bonded onto another cured element (the skin (3)) with multiple thickness changes, using a rigid tooling made of invar used for support and positioning during the curing process. Each element is made with preimpregnated material using automated tape laying. The preform of the beams (J-section) is obtained by hot forming of flat

laminates. The final curing and bonding to the precured skin (co-bonding) is performed using a direct vacuum bag in an autoclave. A flat development of the vacuum bag is performed, it is traced with a numerical control machine and it is made prior to being placed on the tool. For large surfaces with difficult access the final adjustment is performed with the tool and the part in a vertical position, due to the ergonomic difficulties involved in working on certain areas of same. The invention is applicable to the field of aeronautics.



**Fig. 1**

**EP 1 231 046 A3**



European Patent  
Office

# EUROPEAN SEARCH REPORT

Application Number  
EP 02 38 0034

DOCUMENTS CONSIDERED TO BE RELEVANT			
Category	Citation of document with indication, where appropriate, of relevant passages	Relevant to claim	CLASSIFICATION OF THE APPLICATION (Int.Cl.7)
P, A	EP 1 134 070 A (CONST AERONAUTICAS SA) 19 September 2001 (2001-09-19) * claims 1,2,4-6,12; figure 1 * * column 3, line 42 - line 44 *	1-15	B29C70/34 B29C70/44 B64C3/18 B64C3/24
A	US 5 242 523 A (GOODNO KENNETH N ET AL) 7 September 1993 (1993-09-07) * claim 1; figures 1-3 *	1-15	
A	US 5 707 576 A (ASHER DON L) 13 January 1998 (1998-01-13) * claim 1; figure 2 *	1-15	
A	FR 2 440 831 A (DASSAULT AVIONS) 6 June 1980 (1980-06-06) * claims 1,3,8; figures 1,4,6 *	1-15	
			TECHNICAL FIELDS SEARCHED (Int.Cl.7)
			B29C B64C
The present search report has been drawn up for all claims			
Place of search		Date of completion of the search	Examiner
THE HAGUE		2 September 2002	Pierre, N
<p>CATEGORY OF CITED DOCUMENTS</p> <p>X : particularly relevant if taken alone  Y : particularly relevant if combined with another document of the same category  A : technological background  O : non-written disclosure  P : intermediate document</p> <p>T : theory or principle underlying the invention  E : earlier patent document, but published on, or after the filing date  D : document cited in the application  I : document cited for other reasons  &amp; : member of the same patent family, corresponding document</p>			

EP 02 38 0034 (2002-09-13)

ANNEX TO THE EUROPEAN SEARCH REPORT  
ON EUROPEAN PATENT APPLICATION NO.

EP 02 38 0034

This annex lists the patent family members relating to the patent documents cited in the above-mentioned European search report. The members are as contained in the European Patent Office EDP file on  
The European Patent Office is in no way liable for these particulars which are merely given for the purpose of information.

02-09-2002

Patent document cited in search report		Publication date	Patent family member(s)		Publication date
EP 1134070	A	19-09-2001	EP	1134070 A1	19-09-2001
US 5242523	A	07-09-1993	NONE		
US 5707576	A	13-01-1998	US	5622733 A	22-04-1997
FR 2440831	A	06-06-1980	FR	2440831 A1	06-06-1980

EPC FORM P-437

For more details about this annex : see Official Journal of the European Patent Office, No. 12/82